

torsion & thermal effects

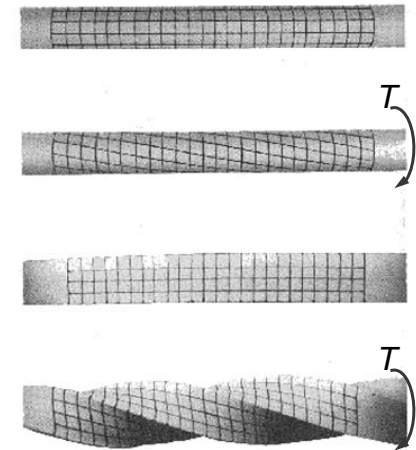
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Torsional Stress & Strain

- can see torsional stresses & twisting of axi-symmetrical cross sections
 - torque
 - remain plane
 - undistorted
 - rotates
- not true for square sections....



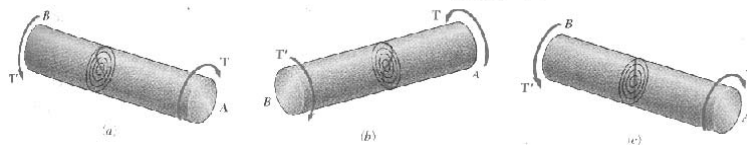
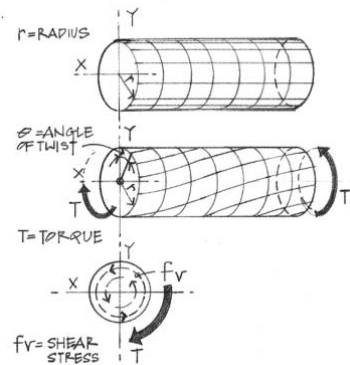
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Shear Stress Distribution

- depend on the deformation
- ϕ = angle of twist
 - measure
- can prove planar section doesn't distort



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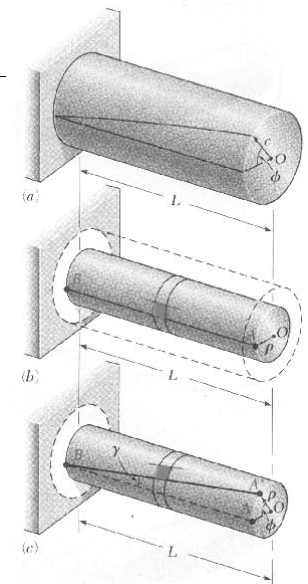
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Shearing Strain

- related to ϕ

$$\gamma = \frac{\rho\phi}{L}$$
- ρ is the radial distance from the centroid to the point under strain
- shear strain varies linearly along the radius: γ_{max} is at outer diameter



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Torsional Stress - Strain

- know $f_v = \tau = G \cdot \gamma$ and $\gamma = \frac{\rho\phi}{L}$
- so $\tau = G \cdot \frac{\rho\phi}{L}$
- where G is the Shear Modulus

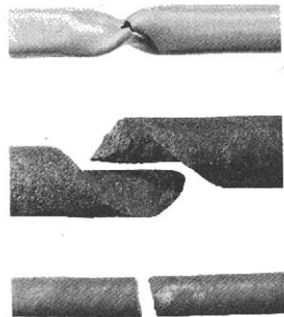
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Shear Stress

- τ_{max} happens at outer diameter
- combined shear and axial stresses
 - maximum shear stress at 45° “twisted” plane



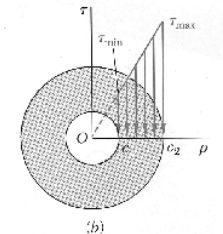
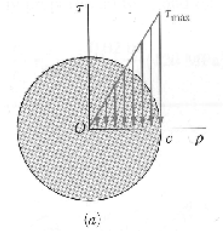
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Torsional Stress - Strain

- from $T = \Sigma \tau(\rho) \Delta A$
- can derive $T = \frac{\tau J}{\rho}$
 - where J is the polar moment of inertia
 - elastic range $\tau = \frac{T\rho}{J}$



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Shear strain

- knowing $\tau = G \cdot \frac{\rho\phi}{L}$ and $\tau = \frac{T\rho}{J}$
- solve: $\phi = \frac{TL}{JG}$
- composite shafts: $\phi = \Sigma_i \frac{T_i L_i}{J_i G_i}$

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Noncircular Shapes

- torsion depends on J
- plane sections don't remain plane
- τ_{max} is still at outer diameter

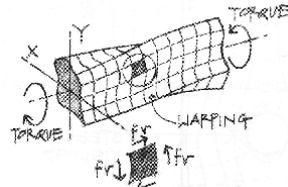


TABLE 3.1. Coefficients for Rectangular Bars in Torsion

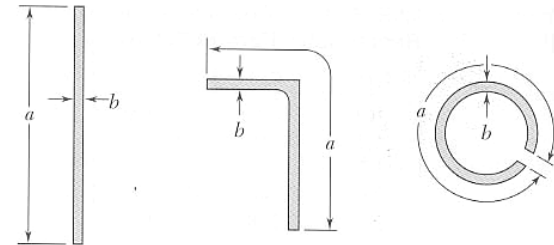
a/b	c_1	c_2
1.0	0.208	0.1406
1.2	0.219	0.1661
1.5	0.231	0.1958
2.0	0.246	0.229
2.5	0.258	0.249
3.0	0.267	0.263
4.0	0.282	0.281
5.0	0.291	0.291
10.0	0.312	0.312
∞	0.333	0.333

$$\tau_{max} = \frac{T}{c_1 ab^2} \quad \phi = \frac{TL}{c_2 ab^3 G}$$

– where a is longer side ($> b$)

Open Thin-Walled Sections

- with very large a/b ratios:



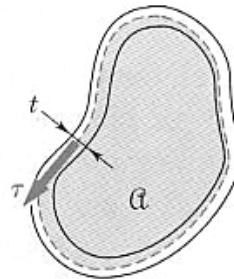
$$\tau_{max} = \frac{T}{\frac{1}{3} ab^2} \quad \phi = \frac{TL}{\frac{1}{3} ab^3 G}$$

Shear Flow in Closed Sections

- q is the internal shear force/unit length

$$\tau = \frac{T}{2t\mathcal{A}}$$

$$\phi = \frac{TL}{4t\mathcal{A}^2} \sum_i \frac{s_i}{t_i}$$



- \mathcal{A} is the area bounded by the centerline
- s_i is the length segment, t_i is the thickness

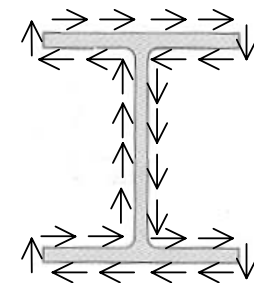
Shear Flow in Open Sections

- each segment has proportion of T with respect to torsional rigidity,

$$\tau_{max} = \frac{Tt_{max}}{\frac{1}{3} \sum b_i t_i^3}$$

- total angle of twist:

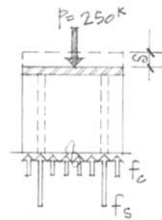
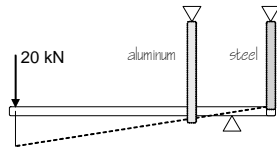
$$\phi = \frac{TL}{\frac{1}{3} G \sum b_i t_i^3}$$



- I beams - web is thicker, so τ_{max} is in web

Deformation Relationships

- physical movement
 - axially (same or zero)
 - rotations from axial changes



- $\delta = \frac{PL}{AE}$ relates δ to P

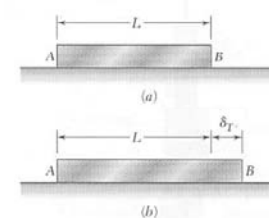
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Deformations from Temperature

- atomic chemistry reacts to changes in energy
- solid materials
 - can contract with decrease in temperature
 - can expand with increase in temperature
- linear change can be measured per degree



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Thermal Deformation

- α - the rate of strain per degree

- UNITS : $\text{1/}^\circ\text{F}$, $\text{1/}^\circ\text{C}$

- length change: $\delta_T = \alpha(\Delta T)L$

- thermal strain: $\epsilon_T = \alpha(\Delta T)$

– no stress when movement allowed

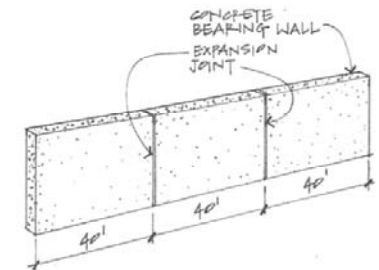
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Coefficients of Thermal Expansion

Material	Coefficients (α) [in./in./°F]
Wood	3.0×10^{-6}
Glass	4.4×10^{-6}
Concrete	5.5×10^{-6}
Cast Iron	5.9×10^{-6}
Steel	6.5×10^{-6}
Wrought Iron	6.7×10^{-6}
Copper	9.3×10^{-6}
Bronze	10.1×10^{-6}
Brass	10.4×10^{-6}
Aluminum	12.8×10^{-6}



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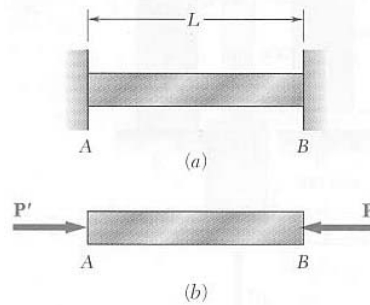
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Stresses and Thermal Strains

- if thermal movement is restrained stresses are induced

- bar pushes on supports
- support pushes back
- reaction causes internal stress

$$f = \frac{P}{A} = \frac{\delta}{L} E$$



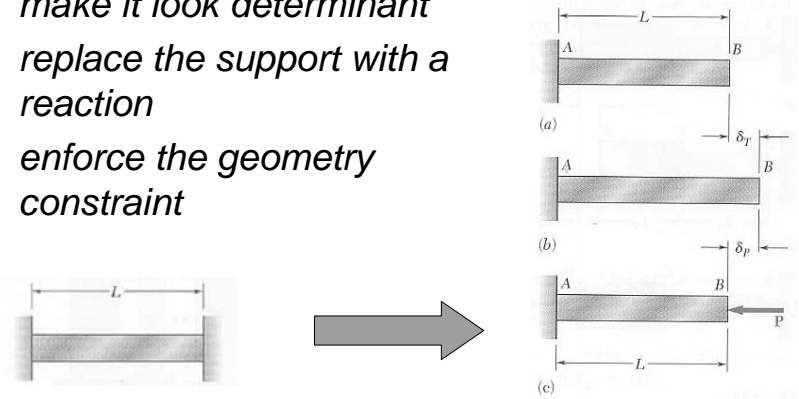
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Superposition Method

- can remove a support to make it look determinant
- replace the support with a reaction
- enforce the geometry constraint



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Superposition Method

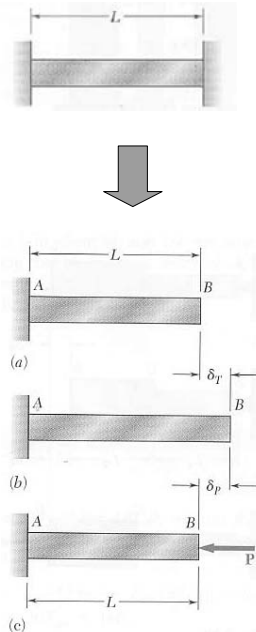
- total length change restrained to zero

constraint: $\delta_P + \delta_T = 0$

$$\delta_P = -\frac{PL}{AE} \quad \delta_T = \alpha(\Delta T)L$$

sub: $-\frac{PL}{AE} + \alpha(\Delta T)L = 0$

$$f = -\frac{P}{A} = -\alpha(\Delta T)E$$



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